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AFSC

AIR PORCE SYSTEMS COMMAND

#### SPACE SYSTEMS PROPULSION TECHNOLOGY

by: Dale Hite

# SPACE SYSTEMS PROPULSION TECHNOLOGY VISION

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#### **NEEDS**

- Full Operational Use of Space Dictates Multi-Purpose Vehicles that Assume:
  - MILITARY RESPONSIVENESS
  - RELIABILITY/MAINTAINABILITY
  - SUPPORTABLE LOGISTICS
  - TECHNOLOGICAL INNOVATION

#### **ELEMENTS**

- SSTO
- ORBIT TRANSFER AND MANEUVERING
- REVOLUTIONARY CAPABILITY

## ORBIT TRANSFER AND MANEUVERING PROPULSION

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#### CHEMICAL PROPULSION

- MODULAR STORABLE PROPULSION 8,000 TO 10,000 LBM TO GEO; IMBEDDED MANEUVERING PROPULSION PACKAGING FLEXIBILITY AND UNLIMITED RESTART BY FY 92
- ROCKET ENGINE MATERIALS 50% WEIGHT REDUCTION WITH ELIMINATION OF CORROSION AND WEAR LIMITS ON RELIABILITY AND COST BY FY 93
- CRYOGENIC ENGINE TECHNOLOGY 13,000 to 15,000 LBM to GEO BY FY 93

# MODULAR/STORABLE ORBIT TRANSFER/MANEUVERING PROPULSION

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#### **FEATURES**

- HIGH PERFORMANCE STORABLE ENGINES (XLR-132)
- MODULAR, COMPACT PROPELLANT FEED SYSTEMS
- PAYLOAD INCREASE TO GEO
  - 45% WITH TITAN IV/IUS
  - MORE THAN 75% WITH SHUTTLE
- 40% More Maneuvering Capability for Survivability
- 50% VOLUME SAVINGS

#### TRANSITION TARGETS

- 10K CLASS ELV UPPER STAGES
- GPS, DSCS, DSP (BLOCK CHANGES)

#### CRYOGENIC OTV PROPULSION

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#### **FEATURES**

- 29% More Payload Volume than Centaur G' (40 FT Payload)
- CAPABLE OF 30 DAY HOLD IN LEO
- ENABLE 40% PAYLOAD INCREASE TO GEO

#### TRANSITION TARGETS

• 13K+ CLASS ELV UPPER STAGES

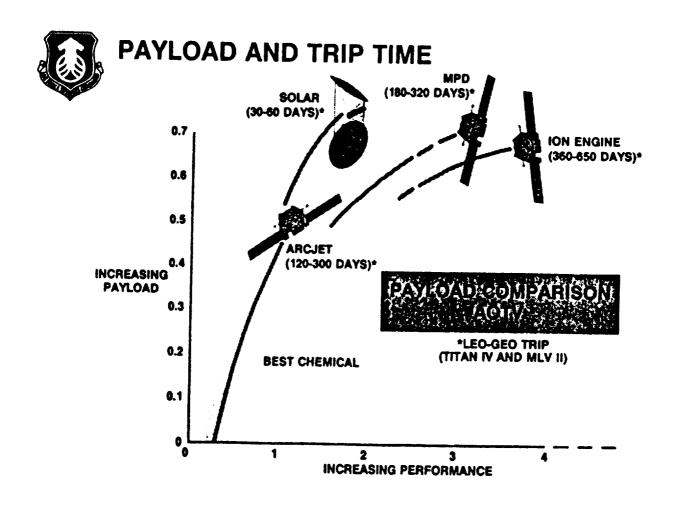
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# ADVANCED SPACE PROPULSION

- ELECTRIC PROPULSION ARCJET 60 TO 100% INCREASE IN PAYLOAD; 12,000 LBS OFF TITAN AND 5,000 LBS OFF ATLAS - BY FY 2000
- ELECTRIC PROPULSION MAGNETOPLASMADYNAMIC THRUSTER -

BY FY 2025

SOLAR PROPULSION - 26,000 TO 45,000 LBS FROM LEO TO GEO, SHUTTLE LAUNCH;
 ELIMINATES SOLAR ARRAY PROBLEMS - BY FY 2010



## ELECTRIC VERSUS CHEMICAL PROPULSION

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#### **ELECTRIC (30KWE ARCJET)**

- PROPELLANT CONSUMPTION 1/2 TO 1/3 THAT OF CHEMICAL
- LARGE POWER SOURCE REQUIREMENT
- FIVE MONTH TRANSFER TIMES

#### **CHEMICAL**

- HIGH PROPELLANT CONSUMPTION
- STORED CHEMICAL ENERGY SOURCE
- SHORT TRANSFER TIMES

### HIGH ENERGY DENSITY MATTER

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## **IDENTIFICATION AND SYNTHESIS**

• IN-HOUSE CENTER OF EXCELLENCE - CONTINUING EFFORT

## PROPELLANT DEVELOPMENT - FY 2005

- DOUBLE PAYLOADS
- REDUCE ROCKET SIZE BY 50%
- SINGLE STAGE TO ORBIT

### OTHER TECHNOLOGY BASE ACTIVITIES

- DOD CRITICAL TECHNOLOGIES LIST
- AFOSR/NC
- NASA LEWIS
- WRDC/POOC

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## NUCLEAR UPPER STAGE OFF ATLAS II

- TECHNOLOGY DEMONSTRATED
  - ACHIEVED ASSEMBLY OF POWER SYSTEM AT SANDIA
  - MEASURED POWER SYSTEM HEATING RATES AT BROOKHAVEN
  - DEMONSTRATED STRUCTURAL INTEGRITY OF POWER SYSTEM AT LOW POWER AND NOMINAL TEMPERATURES
- PAYOFF
  - 400% INCREASE OVER CENTAUR G OFF ATLAS II

### SPACE LAUNCH PROPULSION

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### SSTO PROPULSION TECHNOLOGY

- ALTITUDE COMPENSATING NOZZLE 15 SECOND ISP GAIN AND 35% PAYLOAD GAIN BY FY 91
- Advanced Materials for NASP Propulsion System \$30M Cost Savings, 2,000 lb
  Weight Savings Eliminate Potential Environmental Concern by FY 92
- Long Life Combustion Chamber 300 Cycles by FY 92
- INJECTOR SPRAY CHARACTERIZATION:
  - Manifold and Orifice Hydraulics by FY 91
  - ATOMIZATION AND MIXING BY FY 92
  - SECONDARY DROPLET BREAKUP BY FY 93
  - SUPERCRITICAL VAPORIZATION BY FY 94

# ADVANCED LAUNCH DEVELOPMENT PROGRAM TECHNOLOGIES AT AL

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#### **FEATURES**

- ALTERNATE PYROTECHNICS
- LOW COST EXPENDABLE CRYOGENIC TANKS
- CLEAN PROPELLANT DEVELOPMENT
- SOLID PROPULSION INTEGRATION AND VERIFICATION
- Non-destructive Evaluation of Solid Rocket Boosters
- LOW COST EXPENDABLE LOX/H2 ENGINE DEVELOPMENT
- ROCKET ENGINE CONDITION MONITORING

### AL SUPPORT TO NASP

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## TECHNICAL MANAGEMENT

- ROCKET PROPULSION TECHNOLOGY
- ONE CO-LOCATED AL ENGINEER WITH JOINT PROGRAM OFFICE

## TECHNICAL SUPPORT

• AIR BREATHING ENGINES

## PLANNING SUPPORT

• FEED SYSTEMS AND TURBOPUMPS

## POTENTIAL SUPPORT

TESTING FACILITIES AND FLIGHT TESTING AT AFFTC

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PRESENTATION 1.3.6

UNMANNED LAUNCH VEHICLES / UPPER STAGES